

IN THE UNITED STATES PATENT AND TRADEMARK OFFICE

In re the Application of

Julian G. BALSDON et al.

Group Art Unit: 3641

Application No.: 09/184,402

Examiner: H. Tudor

Filed: October 28, 1998

Docket No.: 101846

For: BLADE TIP CLEARANCE SYSTEM

AMENDMENT UNDER 37 C.F.R. §1.114

RECEIVED

Director of the U.S. Patent and Trademark Office
Washington, D.C. 20231

OCT - 4 2001

LICENSING & REVIEW

Sir:

In reply to the Office Action mailed April 12, 2001, please amend the above-identified application as follows:

IN THE SPECIFICATION:

Please amend the specification as follows:

Page 3, lines 10-21, delete current paragraph and insert therefor:

Referring now to Figure 1, there is shown a radial view through part of the first, high pressure turbine stage of a bypass gas turbine aeroengine. A section of a generally cylindrical engine outer casing is indicated schematically at 2, and an adjacent section of a concentric inner casing, likewise schematically, at 4. An annular space 6 between the outer and inner casings 2,4 constitutes the engine bypass duct. On the left (upstream) side of Figure 1 is shown part of an upstream nozzle guide vane 18 extending radially across a hot gas path 3 between an outer vane platform 16 and a concentric inner vane platform (not shown). As will be understood, the illustrated guide vane 18 is one of a series of guide vanes extending radially between the concentric vane platforms and which together with the platforms form the outlet